



TYPE CERTIFICATE DATA SHEET

N° EASA.R.125

for

SA 341

Type Certificate Holder

Airbus Helicopters

Aéroport International Marseille – Provence

13725 Marignane CEDEX

France

For Models: SA 341 G, SA 342 J



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SECTION 1: SA 341 G**I. General**

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|-----|---|--|
| 1. | Type/ Model/ Variant | |
| 1.1 | Type | SA 341 |
| 1.2 | Model | SA 341 G |
| 1.3 | Variant | - - - |
| 2. | Airworthiness Category | Small Rotorcraft |
| 3. | Manufacturer | Airbus Helicopters
Aéroport International Marseille – Provence
13725 Marignane CEDEX, France |
| 4. | Type Certification Application Date to DGAC | not recorded |
| 5. | State of Design Authority | EASA
(pre EASA: DGAC, France) |
| 6. | Type Certificate Date by DGAC FR | 7 June 1972 |
| 7. | Type Certificate n° | DGAC FR: n° 66
EASA: EASA.R.125 |
| 8. | Type Certificate Data Sheet n° | n° 136 (until issue 4, dated March 1993)
EASA.R.123 (since 7 January 2014) |
| 9. | EASA Type Certification Date | 28 September 2003,
in accordance with CR (EU) 1702/2003, Article 2, 3., (a),
(i), 2 nd bullet, 1 st indented bullet. |

II. Certification Basis

- | | | |
|-----|--|--|
| 1. | Reference Date for determining the applicable requirements | not recorded |
| 2. | Airworthiness Requirements | FAR 27, edition dated 1 February 1965 (Amdt. 27-1 to 27-4 included) with the SGAC additional Special Conditions notified with letter SGAC n° 4986, dated 2 September 1971).
For IFR version, it has been verified that the aircraft equipped according to Aerospatiale drawing ref 341 A/MR 0345 complies with document FAA EU 100 "Acceptable criteria for compliance with FAR 27.141 and 29.141 Instrument flight", dated 15 February 1971. |
| 3. | Special Conditions | Refer to §1 certification basis (see II.2) |
| 4. | Exemptions | none |
| 5. | Deviations | none |
| 6. | Equivalent Safety Findings | none |
| 7. | Requirements elected to comply | none |
| 8. | Environmental Protection Requirements | |
| 8.1 | Noise Requirements | Complies with the essential requirements by virtue of early TC date, see also TCDSN N° EASA.R.125 |
| 8.2 | Emission Requirements | n/a |
| 9. | Operational Suitability Data (OSD) | Not required for rotorcraft that are no longer in production. CR (EU) 748/2012, as amended by CR (EU) |



69/2014 does not require OSD elements for this model
(see Article 7a, 1.).

III. Technical Characteristics and Operational Limitations

1. Type Design Definition SA 341 G basic SA 341 G definition
2. Description
 - Main rotor: three-bladed main rotor
 - Tail rotor: 'Fenestron' type tail rotor
 - Fuselage: airframe of conventional structure
 - Landing gear: skids
 - Powerplant: single turbine
3. Equipment
 - As per compliance with applicable airworthiness requirements defined here above and referenced within approved Rotorcraft Flight Manual.
4. Dimensions
 - 4.1 Fuselage
 - Length: 9.53 m (31.27 ft), or,
 - Width: 2.04 m (6.69 ft) with narrow pads gear
 - Height: 3.19 m (10.47 ft)
 - 4.2 Main Rotor
 - Diameter: 10.50 m (34.45 ft)
 - 4.3 Tail Rotor
 - Diameter: 0.70 m (2.3 ft) 'Fenestron'
5. Engine
 - 5.1 Model
 - SAFRAN Helicopter Engines (Turbomeca)
 - 1 x Model Astazou III A
 - 5.2 Type Certificate
 - EASA TC/TCDS n°: EASA.E.071
 - (DGAC-FR TC/TCDS n°: M6)
 - 5.3 Limitations

5.3.1 Installed Engine Limitations

	PWR [kW]	Gas generator [min ⁻¹]	Temperature T4 [°C]
Max rpm	---	43 500 ¹⁾	---
Max PWR	440	---	---
Max T4 at engine start		---	T4 max + 150
Max T4		---	550
Max T4 continuous	---	---	550

Note: ¹⁾ rpm ±400 rpm (transient loading ±1 500 rpm allowed)

5.3.2 Transmission Torque Limits Refer to approved RFM

6. Fluids (Fuel/ Oil/ Additives)
 - 6.1 Fuel
 - Refer to approved RFM
 - 6.2 Oil
 - Refer to approved RFM for engine and gearboxes
 - 6.3 Additives
 - Refer to approved RFM
 - 6.4 Hydraulic
 - Refer to approved RFM
7. Fluid capacities
 - 7.1 Fuel
 - Total fuel tank capacity: 457 litres (120.7 US gal)
 - Usable fuel: 455 litres (120.2 US gal)
 - Non-usable fuel: 2 litres (0.5 US gal)



- 7.2 Oil
- Engine: 9.2 litres (2.4 US gal)
Main Gear Box: 3.5 litres (0.9 US gal)
Tail Gear Box: 0.3 litres (0.08 US gal)
- 7.3 Coolant System Capacity n/a
8. Air Speed Limitations
- VFR flight:
V_{NE} 310 km/h (167 kt) for 0 m < Z_p < 500m and a decrease of 25 km/h (13.5 kt) must be applied for each 3 280 ft (1 000 m) increase of Z_p.
IFR flight:
V_{NE} corresponding to the speed obtained at level flight for the 2nd pitch stop.
9. Rotor Speed Limitations
- Maximum 420 rpm
Minimum 280 rpm
Max continuous 362 rpm
10. Maximum Operating Altitude and Temperature
- 10.1 Altitude
- 19 685 ft (6 000 m) PA for VFR flight
10 825 ft (3 300 m) PA for IFR flight
- 10.2 Temperature
- Min: -50°C for VFR flight
+2°C for IFR flight
Max: +45°C for -500 m < Z_p < +500 m
and a decrease of 10°C for each Z_p = 1 000 m up to 6 000 m
11. Operating Limitations
- VFR day and night
IFR
12. Maximum Mass
- TKOF/LDG: 1 800 kg (3 968 lb)
13. Centre of Gravity Range
- | Longitudinal | Limit from reference point [mm] | |
|--------------|---------------------------------|-------|
| VFR flight | 2 800 | 3 140 |
| IFR flight | 2 800 | 3 070 |
| Lateral | Limit from reference point [mm] | |
| LH | 153 | |
| RH | 135 | |
14. Datum
- Longitudinal:
3 000 mm (9.84 ft) forward of main rotor centre line
Lateral: aircraft symmetry plane
15. Levelling Means
- 3 levelling legs on the floor
16. Minimum Flight Crew
- VFR/IFR: 1 pilot (RH seat at STA +1 440 mm)
17. Maximum Passenger Seating Capacity
- Four
1 in LH seat at STA +1 440
3 in rear seats at STA +2 290
18. Passenger Emergency Exit
- Refer to approved RFM
19. Maximum Baggage/ Cargo Loads
- The floor of the cabin and the baggage hold has the structural strength necessary for a load uniformly distributed of 610 daN/m.²
20. Rotor Blade Control Movement
- For rigging information refer to the Maintenance Manual



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|--------------------------------|--|
| 21. Auxiliary Power Unit (APU) | n/a |
| 22. Life-limited Parts | The periods specified in the latest approved revision of the Airworthiness Limitations section of the Maintenance Manual must not be exceeded. |

IV. Operating and Service Instructions

- | | |
|--|---|
| 1. Flight Manual | SA 341 G Flight Manual, original edition approved by DGAC, or later DGAC-FR or EASA approved revision. |
| 2. Maintenance Manual | SA 341 G Maintenance Manual |
| 3. Structural Repair Manual | not recorded |
| 4. Weight and Balance Manual | not recorded |
| 5. Illustrated Parts Catalogue | not recorded |
| 6. Miscellaneous Manuals | not recorded |
| 7. Service Letters and Service Bulletins | As published by Aérospatiale, Eurocopter or Airbus Helicopters |
| 8. Required Equipment | As per compliance with applicable requirements and in accordance with the original Type Design standard; refer to approved RFM. |

V. Notes

1. Manufacturer's eligible serial numbers:
Each GAZELLE helicopter manufactured by Aérospatiale and complying with SA 341 G type.
2. The certified "optional" installations are each approved independently of the basic helicopter and an approved Flight Manual Supplement is associated to each optional installation, if necessary.
3. Commercial designation:
GAZELLE
4. SA 341 helicopters manufactured by other European aircraft manufacturers under licence agreement are not covered by Type Certificate EASA.R.125.

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SECTION 2: SA 342 J**I. General**

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|-----|---|--|
| 1. | Type/ Model/ Variant | |
| 1.1 | Type | SA 342 |
| 1.2 | Model | SA 342 J |
| 1.3 | Variant | - - - |
| 2. | Airworthiness Category | Small Rotorcraft |
| 3. | Manufacturer | Airbus Helicopters
Aéroport International Marseille – Provence
13725 Marignane CEDEX, France |
| 4. | Type Certification Application Date to DGAC | not recorded |
| 5. | State of Design Authority | EASA |
| 6. | Type Certificate Date by DGAC FR | 27 April 1976 |
| 7. | Type Certificate n° | DGAC FR: n° 66
EASA: EASA.R.125 |
| 8. | Type Certificate Data Sheet n° | n° 136 (until issue 4, dated March 1993)
EASA.R.123 (since 7 January 2014) |
| 9. | EASA Type Certification Date | 28 September 2003,
in accordance with CR (EU) 1702/2003, Article 2, 3., (a),
(i), 2 nd bullet, 1 st indented bullet. |

II. Certification Basis

- | | | |
|-----|--|--|
| 1. | Reference Date for determining the applicable requirements | not recorded |
| 2. | Airworthiness Requirements | FAR 27, edition dated 1 February 1965 (Amdt. 27-1 to 27-4 included) with the SGAC additional Special Conditions notified with letter SGAC n° 4986, dated 2 September 1971).
For IFR version, it has been verified that the aircraft equipped according to Aerospatiale drawing ref 341 A/MR 0345 complies with document FAA EU 100 "Acceptable criteria for compliance with FAR 27.141 and 29.141 Instrument flight", dated 15 February 1971. |
| 3. | Special Conditions | Refer to §1 certification basis (see II.2) |
| 4. | Exemptions | none |
| 5. | Deviations | none |
| 6. | Equivalent Safety Findings | none |
| 7. | Requirements elected to comply | none |
| 8. | Environmental Protection Requirements | |
| 8.1 | Noise Requirements | Complies with the essential requirements by virtue of early TC date, see also TCDSN N° EASA.R.125 |
| 8.2 | Emission Requirements | n/a |
| 9. | Operational Suitability Data (OSD) | Not required for rotorcraft that are no longer in production. CR (EU) 748/2012, as amended by CR (EU) 69/2014 does not require OSD elements for this model |



(see Article 7a, 1.).

III. Technical Characteristics and Operational Limitations

1. Type Design Definition SA 342 J basic SA 342 J definition
2. Description
 - Main rotor: three-bladed main rotor
 - Tail rotor: 'Fenestron' type tail rotor
 - Fuselage: airframe of conventional structure
 - Landing gear: skids
 - Powerplant: single turbine
3. Equipment
 - As per compliance with applicable airworthiness requirements defined here above and referenced within approved Rotorcraft Flight Manual.
4. Dimensions
 - 4.1 Fuselage
 - Length: 9.53 m (31.27 ft), or,
 - Width: 2.04 m (6.69 ft) with narrow pads gear
 - Height: 3.19 m (10.47 ft)
 - 4.2 Main Rotor
 - Diameter: 10.50 m (34.45 ft)
 - 4.3 Tail Rotor
 - Diameter: 0.70 m (2.3 ft) 'Fenestron'
5. Engine
 - 5.1 Model
 - SAFRAN Helicopter Engines (Turbomeca)
 - 1 x Model Astazou XIV H
 - 5.2 Type Certificate
 - EASA TC/TCDS n°: EASA.E.075
 - (DGAC-FR TC/TCDS n°: M3)
 - 5.3 Limitations

5.3.1 Installed Engine Limitations

	PWR [kW]	Gas generator [min ⁻¹]	Temperature T4 [°C]
Max rpm	---	43 000 ¹⁾	---
Max PWR	440	---	---
Max T4 at engine start (5 sec)		---	700
Max T4		---	550
Max T4 continuous	---	---	550

Note: ¹⁾ rpm ±200 rpm (transient loading ±1 500 rpm allowed)**5.3.2 Transmission Torque Limits** Refer to approved RFM

6. Fluids (Fuel/ Oil/ Additives)
 - 6.1 Fuel
 - Refer to approved RFM
 - 6.2 Oil
 - Refer to approved RFM for engine and gearboxes
 - 6.3 Additives
 - Refer to approved RFM
 - 6.4 Hydraulic
 - Refer to approved RFM
7. Fluid capacities
 - 7.1 Fuel
 - Total fuel tank capacity: 457 litres (120.7 US gal)
 - Usable fuel: 455 litres (120.2 US gal)
 - Non-usable fuel: 2 litres (0.5 US gal)



- 7.2 Oil
- Engine: 9.2 litres (2.4 US gal)
Main Gear Box: 3.5 litres (0.9 US gal)
Tail Gear Box: 0.3 litres (0.08 US gal)
- 7.3 Coolant System Capacity
- n/a
8. Air Speed Limitations
- VFR flight:
V_{NE} 310 km/h (167 kt) for 0 m < Z_p < 500m and a decrease of 25 km/h (13.5 kt) must be applied for each 3 280 ft (1 000 m) increase of Z_p.
IFR flight:
V_{NE} corresponding to the speed obtained at level flight for the 2nd pitch stop.
9. Rotor Speed Limitations
- Maximum regulated: 387 rpm (powered flight)
Maximum: 430 rpm (Autorotation)
415 rpm for t < 15 °C and Z_p < 9 842 ft (3 000 m)
400 rpm for t < -30 °C
Minimum: 310 rpm for Z_p < 3 280 ft (1 000 m)
for Z_p > 3 280 ft (1 000 m) add 10 rpm for each 3 280 ft (1 000 m) increase of Z_p.
10. Maximum Operating Altitude and Temperature
- 10.1 Altitude
- 19 685 ft (6 000 m) PA for VFR flight
10 825 ft (3 300 m) PA for IFR flight
- 10.2 Temperature
- Min: -40°C for VFR flight
+2°C for IFR flight
Max: +50°C for -500 m < Z_p < +500 m
and a decrease of 10°C for each Z_p = 1 000 m up to 6 000 m
11. Operating Limitations
- VFR day and night
IFR
12. Maximum Mass
- TKOF/LDG: 1 900 kg (4 189 lb)
13. Centre of Gravity Range
- | Longitudinal | Limit from reference point [mm] | |
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| VFR flight | 2 800 | 3 140 |
| IFR flight | 2 800 | 3 070 |
-
- | Lateral | Limit from reference point [mm] | |
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| LH | 153 | |
| RH | 135 | |
14. Datum
- Longitudinal:
3 000 mm (9.84 ft) forward of main rotor centre line
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16. Minimum Flight Crew
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17. Maximum Passenger Seating Capacity
- Four
1 in LH seat at STA +1 440
3 in rear seats at STA +2 290
18. Passenger Emergency Exit
- Refer to approved RFM



- | | |
|----------------------------------|--|
| 19. Maximum Baggage/ Cargo Loads | The floor of the cabin and the baggage hold has the structural strength necessary for a load uniformly distributed of 610 daN/m. ² |
| 20. Rotor Blade Control Movement | For rigging information refer to the Maintenance Manual |
| 21. Auxiliary Power Unit (APU) | n/a |
| 22. Life-limited Parts | The periods specified in the latest approved revision of the Airworthiness Limitations section of the Maintenance Manual must not be exceeded. |

IV. Operating and Service Instructions

- | | |
|--|---|
| 1. Flight Manual | SA 342 J Flight Manual, original edition approved by DGAC, or later DGAC-FR or EASA approved revision. |
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| 3. Structural Repair Manual | not recorded |
| 4. Weight and Balance Manual | not recorded |
| 5. Illustrated Parts Catalogue | not recorded |
| 6. Miscellaneous Manuals | not recorded |
| 7. Service Letters and Service Bulletins | As published by Aérospatiale, Eurocopter or Airbus Helicopters |
| 8. Required Equipment | As per compliance with applicable requirements and in accordance with the original Type Design standard; refer to approved RFM. |

V. Notes

1. Manufacturer's eligible serial numbers:
Each GAZELLE helicopter manufactured by Aérospatiale and complying with SA 342 J type.
2. The certified "optional" installations are each approved independently of the basic helicopter and an approved Flight Manual Supplement is associated to each optional installation, if necessary.
3. Commercial designation:
GAZELLE
4. SA 342 helicopters manufactured by other European aircraft manufacturers under licence agreement are not covered by Type Certificate EASA.R.125.

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SECTION 3: ADMINISTRATIVE**I. Acronyms and Abbreviations**

°C	Degree Celsius	RFM	Rotorcraft Flight Manual
C.G.	Centre of Gravity	rpm	Rounds per minute
CR	(European) Commission Regulation	s/n	Serial Number
EU	European Union	sec	Seconds
LDG	Landing	STA	Station
Max	Maximum	TC	Type Certificate
n/a	not applicable	TCDS	Type Certificate Data Sheet
n°	Number	TKOF	Take-Off
OSD	Operational Suitability Data	VFR	Visual Flight Rules
PA	Pressure Altitude	V _{NE}	Never Exceed Speed
PWR	Power		

II. Type Certificate Holder Record

Type Certificate Holder	Period
Aérospatiale 37, Boulevard de Montmorency 75781 Paris CEDEX 16, France	From 1 January 1970 until 31 December 1991
Eurocopter France Aéroport International Marseille – Provence 13725 Marignane CEDEX, France	From 1 January 1992 until 30 May 1997
Eurocopter Aéroport International Marseille – Provence 13725 Marignane CEDEX, France	From 1 June 1997 until 6 January 2014
Airbus Helicopters Aéroport International Marseille – Provence 13725 Marignane CEDEX, France	Since 7 January 2014

III. Change Record

Issue	Date	Changes	TC issue
Issue 01	27 Jan 2010	Initial issue of EASA TCDS	Re-issued on 27 January 2010
---	7 Jan 2014	The company name has been changed to AIRBUS HELICOPTERS	Re-issued on 7 January 2014
Issue 02	14 Feb 2017	New TCDS template, reference to OSD, minor editorial corrections	---

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